

IN THE CLAIMS

Please cancel claims 1-22, 35 and 36 without prejudice or disclaimer, add new claims 50 and 51 and amend the remaining claims as indicated:

1-22. (Cancelled)

23. (Currently Amended) A method of producing a balanced Al-Cu-Mg-Si alloy ~~according to any one of claim 1 and~~ having a high toughness, good strength levels and an improved fatigue crack growth resistance, comprising the steps of:

a) casting an ingot comprising the following composition (in weight percent):

Cu: 3.6 - 4.9

Mg: 1.0 - 1.8

~~Mn: < 0.30~~

Si: 0.10 - 0.40

Zr:  $\leq$  0.15

Cr:  $\leq$  0.15

Fe:  $\leq$  0.10,

the balance essentially aluminum and incidental elements and impurities, wherein the ingot is cast by semi-continuous direct chill (DC) casting,

b) homogenizing and/or pre-heating the ingot after the casting step,

c) hot rolling the homogenized and/or pre-heated ingot and optionally cold rolling into a rolled product,

d) solution heat treating the hot rolled product,

e) quenching the solution heat treated product,

f) stretching the quenched product, and

g) naturally ageing the stretched, rolled and heat-treated product.

24. (Original) Method according to claim 23, wherein, after hot rolling the ingot, annealing and/or reheating the hot rolled ingot and further hot rolling the rolled ingot.

25. (Original) Method according to claim 23, wherein said hot rolled ingot is inter-annealed before and/or during cold rolling.

26. (Original) Method according to claim 23, wherein said rolled and heat-treated product is stretched in a range of up to 3% and naturally aged for more than 10 days.

27. (Original) Method according to claim 23, wherein the rolled and heat-treated product is stretched in a range of 1 to 2%.

28. (Original) Method according to claim 23, wherein the rolled and heat-treated product is, after stretching, naturally aged for a period in a range of 10 to 20 days.

29. (Original) Method according to claim 23, wherein the alloy product has been processed to an alloy product in a T3 or T351 temper condition.

30. (Original) Method according to claim 23, wherein the alloy product has been processed to a sheet product having a final thickness in a range of 2.0 to 12 mm.

31. (Original) Method according to claim 23, wherein the alloy product has been processed to a sheet product having a final thickness in a range of 25 to 50 mm.

32. (Original) Method according to claim 23, wherein the alloy product has been processed to a structural member of an aircraft or spaceship.

33. (Original) Method according to claim 23, wherein the alloy product has been processed to a fuselage skin of an aircraft.

34. (Original) Method according to claim 23, wherein the alloy product has been processed to a lower-wing member of an aircraft.

35-36. (Cancelled)

37. (Original) Method according to claim 23, wherein the alloy product is substantially Mn-free.

38. (Original) Method according to claim 23, wherein the amount (in weight %) of Cu in the alloy is in a range of 3.9 to 4.6%.

39. (Original) Method according to claim 23, wherein the amount (in weight %) of Cu in the alloy is in a range of 4.3 to 4.5%

40. (Original) Method according to claim 23, wherein the amount (in weight %) of Mg in the alloy is in a range of 1.2 to 1.7%.

41. (Original) Method according to claim 23, wherein the amount (in weight %) of Mg in the alloy is in a range of 1.5 to 1.7%.

42. (Original) Method according to claim 23, wherein the amount (in weight %) of Si in the alloy is in a range of 0.15 to 0.35 %.

43. (Original) Method according to claim 23, wherein the amount (in weight %) of Si in the alloy is in a range of 0.23 to 0.30%.

44. (Original) Method according to claim 23, wherein the amount (in weight %) of Si in the alloy is in a range of 0.23 to 0.28%.

45. (Original) Method according to claim 23, wherein said alloy further comprises one or more of the elements Zn, Ag, Hf, V, Sc, Ti or Li, the total amount less than 1.00 (in weight %).

46. (Original) Method according to claim 23, wherein the alloy product has been processed to a product having a fatigue crack growth rate of less than 0.001 mm/cycles at  $\Delta K=20$  MPa $\sqrt{m}$  when tested according to ASTM-E647 on 80 mm wide M(T) panels at R=0.1 at constant load and at a frequency of 8 Hz.

47. (Original) Method according to claim 23, wherein the alloy product has been processed to a product having a fatigue crack growth rate of less than 0.01 mm/cycles at  $\Delta K=20$  MPa $\sqrt{m}$  when tested according to ASTM-E647 on 80 mm wide M(T) panels at R=0.1 at constant load and at a frequency of 8 Hz.

48. (Original) Method according to claim 23, wherein the alloy product has been processed to a product having a tensile yield strength of not less than 310 MPa in the L-direction.

49. (Original) Method according to claim 23, wherein the alloy product has been processed to a product having an ultimate tensile strength in the L-direction of not less than 430 MPa.

50. (New) Method according to claim 23, wherein the amount of Fe in the alloy is 0.06-0.10%.

51. (New) Method according to claim 23, wherein the alloy comprises Fe.